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PATENT APPLICATION TRANSMITTAL LETTER

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Transmitted herewith for filing is the patent application of
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(Inventor)

For -----METHODS AND APPARATUS FOR ROTOR OVER-SPEED PROTECTION-----

(Title of Invention)

Enclosed are:

- ☒ 4 sheets of ☒ formal drawing. ☐ Please file concurrently with _____
☐ informal _____
☐ an assignment of the invention to **GENERAL ELECTRIC COMPANY**

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METHODS AND APPARATUS FOR ROTOR OVER-SPEED PROTECTION

BACKGROUND OF THE INVENTION

This invention relates generally to gas turbine engine rotors and, more particularly, to fuel system interfaces used to prevent rotor over-speeds.

Gas turbine engines typically include over-speed protection systems that provide rotor over-speed protection. Typically the over-speed protection systems either hold rotor speed below critical rotor speeds or shut off fuel flow with an electronic signal. One type of known protection system receives over-speed signals from mechanical speed sensors. The mechanical speed sensors include rotating flyweight sensing systems that indicate an over-speed condition as a result of the rotor rotating above the normal operational maximum speeds, yet below the structural failure limits. The flyweight sensing systems are hydraulically coupled to a bypass valve, such that as the flyweights are forced centrifugally outward, the bypass valve is operated to reduce a differential pressure across a fuel metering valve at a fixed value. Thus, fuel flow to the engine is controlled by the flow area of the metering valve.

Other types of known over-speed protection systems receive over-speed signal information from electronic control sensors. Additional mechanical control sensors derive over-speed conditions as a function of an engine fuel flow pressure. Such systems provide for rapid fuel shutoff and resulting engine shutdown if engine speed exceeds the normal maximum value.

Because the protection systems are non-redundant, over-speed protection system architecture varies depending on whether the protection system receives a signal from an electrically or mechanically controlled system. Because of the various system architectures, an elapsed time for the protection system to react to an over-speed condition may vary depending on the type of sensing system used to detect the over-speed condition and a location of the sensing system sensors relative to the protection system.

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BRIEF SUMMARY OF THE INVENTION

In an exemplary embodiment, a fuel system interface for a gas turbine engine prevents a rotor from over-speeding and receives an over-speed signal from either an electrically-originated or a mechanically-originated over-speed sensing system. In the exemplary embodiment, the engine includes a fuel system interface including a fuel shutoff shuttle valve, a fuel shutoff valve, a fuel bypass valve, a plurality of selector valves, an aircraft control shutoff valve, and an independent speed sensing system. The fuel shutoff shuttle valve controls fuel flow to the fuel bypass valve and the fuel shutoff valve, and is coupled to the selector valves. The independent sensing system and the aircraft control shutoff valve are coupled to the selector valves. The aircraft control valve and the shutoff shuttle valve provide parallel redundant fuel cutoff devices that each use selector valves that are employed for normal engine operation.

The fuel system interface includes a shutoff shuttle valve that, in operation, stops fuel flow to the engine to prevent the rotor from over-speeding when an over-speed condition is sensed by the independent sensing system. Because the fuel system interface uses selector valves, rapid signal selection is facilitated within the fuel system interface regardless of a physical proximity of the over-speed sensing system to the fuel system interface shutoff shuttle valve. Furthermore priority logic utilized by the over-speed system provides that when the fuel system interface is activated during over-speed conditions, fuel flow can only be initiated when the over-speed signal is removed.

BRIEF DESCRIPTION OF THE DRAWINGS

Figure 1 is a schematic illustration of a gas turbine engine;

Figure 2 is a schematic illustration of a portion of the engine shown in Figure 1 including a fuel system interface;

Figure 3 is a priority logic table used with the fuel system interface shown in Figure 2;

Figure 4 is a schematic illustration of an alternative embodiment of the fuel system interface shown in Figure 2; and

Figure 5 is a schematic illustration of another alternative embodiment of the fuel system interface shown in Figure 2.

DETAILED DESCRIPTION OF THE INVENTION

5 Figure 1 is a schematic illustration of a gas turbine engine 10 including a low pressure compressor 12, a high pressure compressor 14, and a combustor 16. Engine 10 also includes a high pressure turbine 18, and a low pressure turbine 20. Compressor 12 and turbine 20 are coupled by a first rotor shaft 24, and compressor 14 and turbine 18 are coupled by a second rotor shaft 26. In one embodiment, engine 10 is a GE90 engine available from General Electric Aircraft Engines, Cincinnati, Ohio.

In operation, air flows through low pressure compressor 12 and compressed air is supplied from low pressure compressor 12 to high pressure compressor 14. Compressed air is then delivered to combustor 16 and airflow from combustor 16 drives turbines 18 and 20.

15 Figure 2 is a schematic illustration of a portion of engine 10 including a fuel system interface 36. Figure 3 is an exemplary embodiment of a priority logic table 38 that may be used with fuel system interface 36. Fuel system interface 36 prevents engine rotors, such as turbines 18 and 20, from over-speeding when an independent speed sensing system 40 determines normal engine operating limits have been exceeded, and provides for normal operation of engine 10.

25 Fuel system interface 36 includes a fuel shutoff valve 44, a fuel bypass valve 46, a fuel shutoff shuttle valve 48, and a plurality of selector valves 50. Fuel system interface supplies fuel to engine 10 where it is mixed with air and ignited, producing hot combustion gases used to drive a rotor, such as turbines 18 and 20 (shown in Figure 1). In one embodiment, fuel bypass valve 46 is a fuel metering valve head regulating valve. Shutoff shuttle valve 48, is connected in flow communication with fuel bypass valve 46 and fuel shutoff valve 44, and operates in

either a "RUN" position or in an "OFF" position to control fuel flow through fuel bypass and fuel shutoff valves 46 and 44, respectively.

Fuel system interface 36 also includes a plurality of fuel circuits 60 including a fuel supply pressure circuit 62 and a fuel return pressure circuit 64. Shutoff shuttle valve 48 is connected in flow communication with fuel circuits 60. Fuel flow supplied through fuel supply circuit 62 and fuel return pressure circuit 64 through shutoff shuttle valve 48 varies depending on whether shutoff shuttle valve 48 is selected to operate in the "RUN" or "OFF" position.

Shutoff shuttle valve 48 is coupled to selector valves 50 and receives a signal 66 from selector valves 50 based on logic defined in priority logic table 38. More specifically, shutoff valve 48 receives a "HI/LO" signal 66 from a first selector valve 67, in which the "HI" signal has been predefined as "RUN" and the "LO" signal has been predefined as "OFF", with respect to selected engine operation. "RUN" and "OFF" are produced based the combination of connections between fuel circuits 60 of shutoff valve 48, fuel shutoff valve 44, and fuel bypass valve 46, as well as a position of fuel bypass valve 46 in response to HI/LO" signal 66 received from selector valves 50.

First selector valve 67 is coupled to a second selector valve 68, and selector valves 50 are pressure sensitive devices that select a lowest of two pressure signals inputted for output from selector valves 50. In the exemplary embodiment shown in Figure 2, selector valves 50 prioritize signal selection 66 from three sources for control of shutoff shuttle valve 48. More specifically, selector valves 50 receive inputs from an aircraft control shutoff valve 70, speed sensing system 40, and a fuel metering system (not shown). In one embodiment, aircraft control shutoff valve 70 is an electro-hydraulic valve.

The fuel metering valve transmits a fuel metering valve interlock signal 72 to selector valves 50. The fuel metering system, speed sensing system 40, and aircraft control shutoff valve 70 transmit a "HI/LO" signal 72, 74, and 76, respectively, and because selector valves 50 select a lowest pressure, any of the fuel

metering system, speed sensing system 40, or aircraft control shutoff valve 70 may transmit a "LO" signal to command an "OFF" position of shutoff shuttle valve 48. Conversely, all three signal inputs, 72, 74, and 76 must be "HI" to command a "RUN" position of shutoff shuttle valve 48.

5 Aircraft control shutoff valve 70 is coupled to fuel circuits 60 and selector valves 50. More specifically, aircraft control shutoff valve 70 is coupled to fuel supply pressure circuit 62 and fuel return pressure circuit 64, such that operating aircraft control shutoff valve 70 provides "HI/LO" signal 76 to second selector valve 68. First selector valve 67 selects a lowest pressure between signal 76 and fuel
10 metering valve interlock signal 72, and transmits a signal 80 to first selector valve 67.

Speed sensing system 40 includes an electro-hydraulic valve 90 coupled to selector valves 50 and fuel circuits 60. More specifically, speed sensing system 40 is coupled to fuel supply pressure circuit 62 and fuel return pressure circuit 64, such that operating valve 90 provides "HI/LO" signal 74 to first selector valve 67.
15 First selector valve 67 selects a lowest pressure between signal 80 transmitted by second selector valve 68 and signal 74 transmitted by electro-hydraulic valve 90, and transmits signal 66 to shutoff shuttle valve 48. Speed sensing system 40 is also coupled to an engine control system (not shown). In one embodiment, speed sensing system 40 is coupled to an engine control system known as a full authority digital
20 electronic control (FADEC) available from General Electric Aircraft Engines, Cincinnati, Ohio.

During operation, fuel system interface 36 provides access to engine fuel shut off valve 44 and fuel bypass valve 46 with external signals to shut off engine fuel flow due to an over-speed condition. More specifically, fuel system interface 36
25 stops engine fuel flow to engine 10 by closing fuel shut off valve 44, while fuel bypass valve 46 is opened to stop metered fuel to engine 10 when speed sensing system 40 senses an over-speed condition or as a result of a "LO" fuel metering valve interlock signal 72 or a "LO" signal 76 transmitted from aircraft control shutoff valve 70.

Speed sensing system electro-hydraulic valve 90 is controlled by the engine control system and latches, as illustrated in Figure 2, when the engine control system signals that a pre-determined rotor speed is exceeded. As a result, shutoff shuttle valve 48 stops fuel flow to engine 10 to prevent the rotor from over-speeding.

5 Because fuel system interface 36 acts on both fuel metering head regulator or fuel bypass valve 46 and normal fuel shut off valve 44, rotor over-speed protection is provided via parallel redundant fuel cut-off devices using fuel metering devices 50 that are used for normal engine operation.

10 Priority logic table 38 illustrates the conditions under which engine fuel flow may be initiated in light of the various combinations of signals affecting speed sensing system 40 ("Overspeed Signal"), aircraft control shutoff valve 70 ("Aircraft Signal") and shutoff shuttle valve 48 ("Normal Engine Control Signal"). More specifically, priority logic table 38 provides that when fuel system interface 36 is activated, as a result of receiving an over-speed indication 74 from speed sensing system 40, fuel flow can only be initiated when the over-speed signal is removed.

15 Figure 4 is a schematic illustration of an alternative embodiment of a fuel system interface 100. Fuel system interface 100 is substantially similar to fuel system interface 36, shown in Figure 2 and components in fuel system interface 100 that are identical to components of fuel system interface 36 are identified in Figure 4 using the same reference numerals used in Figure 2. Fuel system interface 100 includes a speed sensing system 102 coupled to selector valves 50 and fuel circuits 60. More specifically, speed sensing system 102 is coupled to fuel supply pressure circuit 62 and fuel return pressure circuit 64, such that speed sensing system 102 may limit or stop fuel flow within a portion of fuel supply pressure and fuel return pressure circuits 20 62 and 64, respectively. Speed sensing system 102 is coupled to a mechanical speed sensor 110 that latches when a pre-determined speed is exceeded. Mechanical speed sensor 110 is known in the art and is shown latched in an over-speed condition.

Figure 5 is a schematic illustration of an alternative embodiment of a fuel system interface 200. Fuel system interface 200 is substantially similar to fuel

system interface 36 (shown in Figure 2) and includes a speed sensing system 202. Speed sensing system 202 is coupled to selector valves 50 and fuel circuits 60. More specifically, speed sensing system 202 is coupled to fuel supply pressure circuit 62 and fuel return pressure circuit 64, such that speed sensing system 202 may limit or
5 stop fuel flow within a portion of fuel supply pressure and fuel return pressure circuits 62 and 64, respectively.

Speed sensing system 202 includes a sensing valve 210, a sensing orifice 212, and a main fuel pump 214. Sensing orifice 212 derives over-speed information from a pressure generated by main fuel pump 214. Sensing orifice 212 is
10 downstream from main fuel pump 214 and receives a total pump fuel flow exiting main fuel pump 214. Because the total pump fuel flow is proportional to engine speed, the sensed pressure derived by sensing orifice 212 increases with engine rotor speed. Sensing valve 210 latches when an over-speed occurs based on over-speed information derived with sensing orifice 212. Sensing valve 210 is shown latched in
15 Figure 5 in an over-speed condition.

The above-described fuel system interface is cost-effective and highly accurate. The fuel system interface facilitates a rapid fuel shutoff to prevent rotor over-speeds from occurring when an independent speed sensing system determines normal rotor operating limits have been exceeded. The interface accommodates over-
20 speed signals that originated from electronic or mechanical speed sensors. As a result, the fuel system interface prevents rotor over-speeds in a cost-effective and reliable manner.

While the invention has been described in terms of various specific embodiments, those skilled in the art will recognize that the invention can be practiced
25 with modification within the spirit and scope of the claims.

WHAT IS CLAIMED IS:

1. A method for assembling a gas turbine engine to prevent rotor over-speeding, said method comprising the steps of:

coupling a fuel system interface to the gas turbine engine such that the fuel system interface receives electrically and mechanically originated over-speed signals inputted from the engine; and

configuring the fuel system interface to stop engine fuel flow in response to the over-speed signals received.

2. A method in accordance with Claim 1 wherein the gas turbine engine includes a fuel metering head regulator and a normal fuel shutoff valve, said step of coupling a fuel system interface further comprises the step of coupling the fuel system interface to the fuel metering head regulator and the normal fuel shutoff valve.

3. A method in accordance with Claim 2 wherein said step of configuring the fuel system interface further comprises the step of configuring the fuel system interface to prevent engine fuel flow to the fuel metering head regulator and the normal fuel shutoff valve when the fuel system interface is activated as a result of an over-speed signal.

4. A method in accordance with Claim 1 wherein said step of coupling a fuel system interface further comprises the step of coupling the fuel system interface to a mechanical speed sensor.

5. A method in accordance with Claim 1 wherein said step of coupling a fuel system interface further comprises the step of coupling the fuel system interface to an engine control system.

6. A fuel system interface for a gas turbine engine including a rotor, said interface configured to receive electrically and mechanically originated over-speed signals from the engine, said interface further configured to stop engine fuel

flow in response to the over-speed signals received to prevent the rotor from over-speeding.

5 7. A fuel system interface in accordance with Claim 6 wherein the engine includes a fuel metering head regulator, said interface coupled to the fuel metering head regulator.

8. A fuel system interface in accordance with Claim 6 wherein the engine includes a normal fuel shutoff valve, said interface coupled to the normal fuel shutoff valve.

10 9. A fuel system interface in accordance with Claim 6 wherein said interface further configured to receive an electrical signal originating from an engine control system.

10. A fuel system interface in accordance with Claim 6 wherein said interface further configured to receive a signal from a mechanical speed sensor.

15 11. A fuel system interface in accordance with Claim 6 wherein the engine includes a fuel metering head regulator and a normal fuel shutoff valve, said interface coupled to the fuel metering head regulator and the normal fuel shutoff valve.

20 12. A fuel system interface in accordance with Claim 11 wherein said interface further configured such that engine fuel flow to the fuel metering head regulator and the normal fuel shutoff valve is prevented when said fuel system interface is activated as a result of an over-speed signal.

13. A gas turbine engine comprising:

a rotor;

25 a fuel delivery system configured to supply fuel to said engine for operating said rotor; and

a fuel system interface coupled to said fuel delivery system and configured to receive a plurality of electrically and mechanically originated over-speed signals from the engine, said interface further configured to stop engine fuel flow in response to the over-speed signals received to prevent said rotor from over-speeding.

14. A gas turbine engine in accordance with Claim 13 wherein said fuel delivery system comprises an engine fuel shut off valve and a fuel metering valve head regulating valve.

15. A gas turbine engine in accordance with Claim 14 wherein said fuel system interface coupled to said engine fuel shut off valve and a fuel metering valve head regulating valve.

16. A gas turbine engine in accordance with Claim 13 wherein said fuel system interface further configured to receive an electrical signal originating from an engine control system.

17. A gas turbine engine in accordance with Claim 13 wherein said fuel system interface further configured to receive a signal from a mechanical speed sensor.

18. A gas turbine engine in accordance with Claim 13 wherein said fuel system interface further configured such that when said fuel system interface activated as a result of sensing an over-speed signal, fuel flow to said engine prevented until the over-speed signal is removed.

METHODS AND APPARATUS FOR ROTOR OVER- SPEED PROTECTION

ABSTRACT OF THE DISCLOSURE

5 A fuel system interface for a gas turbine engine prevents a rotor from over-speeding and receives either an electrically-originated or a mechanically-originated over-speed signal. In the exemplary embodiment, the fuel system interface is coupled to an gas turbine engine fuel shut off valve and to a fuel metering valve to shut off engine fuel after receiving an over-speed condition. As a result, because the fuel system interface acts on both the shut off valve and the fuel metering valve, rotor over-speed protection is facilitated with parallel redundant fuel cutoff devices.

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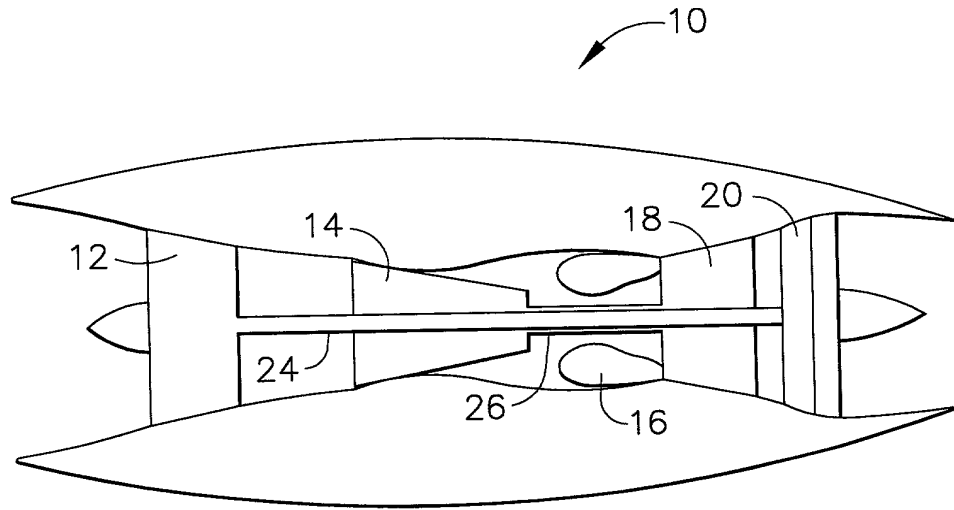
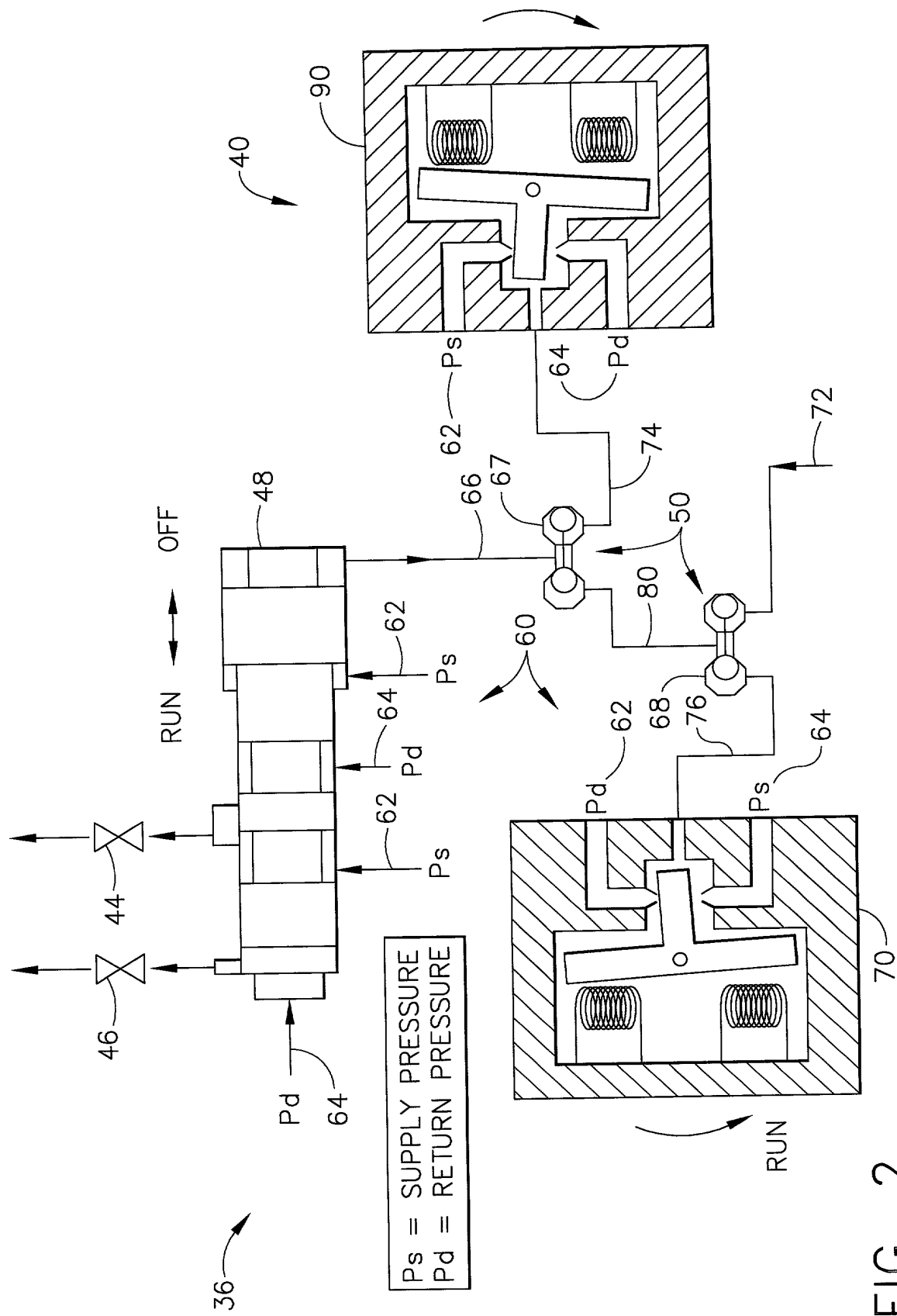


FIG. 1

OVERSPEED SIGNAL	AIRCRAFT SIGNAL	NORMAL ENGINE CONTROL SIGNAL	ENGINE FUEL FLOW
NO	OFF	OFF	NO
NO	OFF	ON	NO
NO	ON	OFF	NO
NO	ON	ON	YES
YES	OFF	OFF	NO
YES	OFF	ON	NO
YES	ON	OFF	NO
YES	ON	ON	NO

FIG. 3



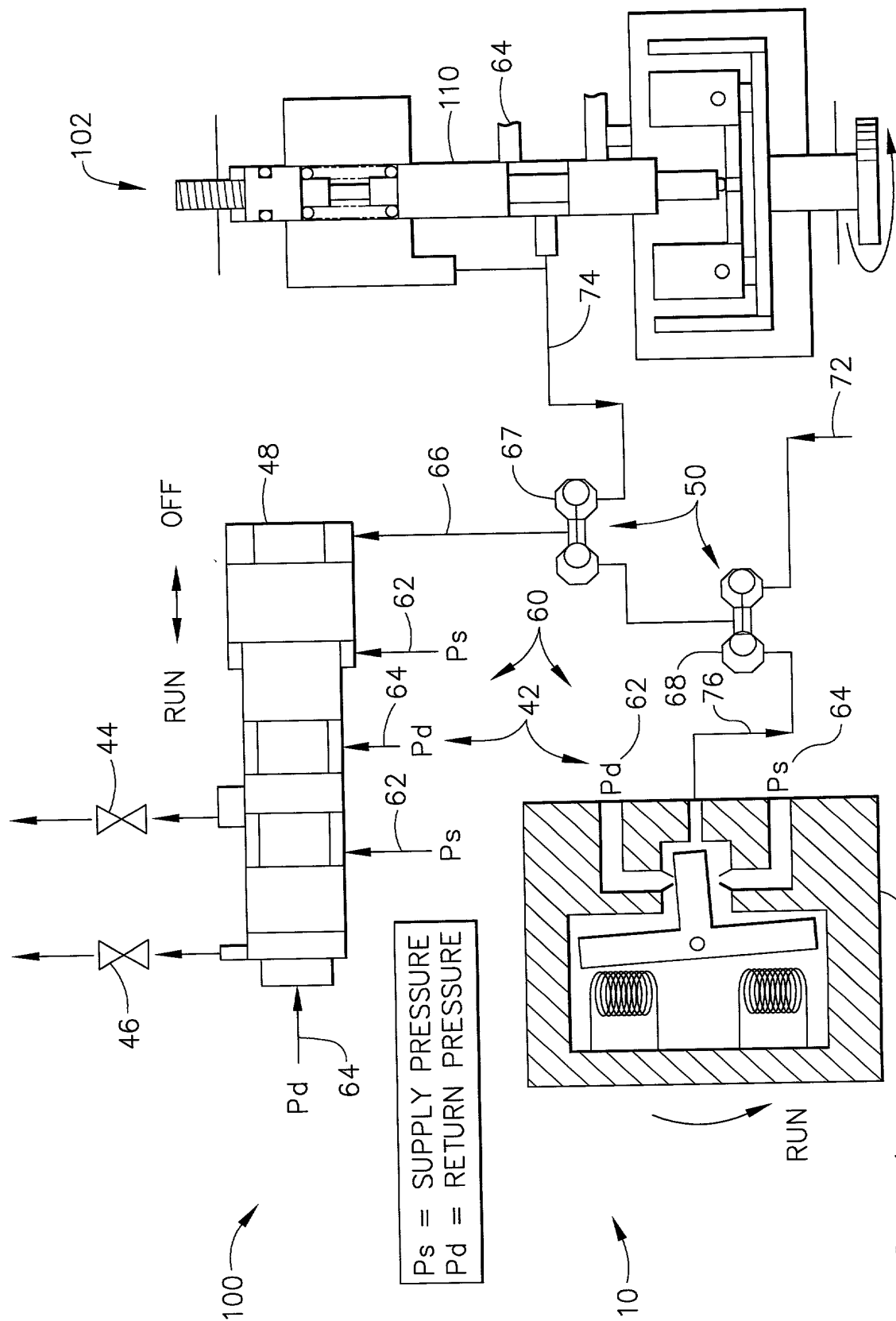
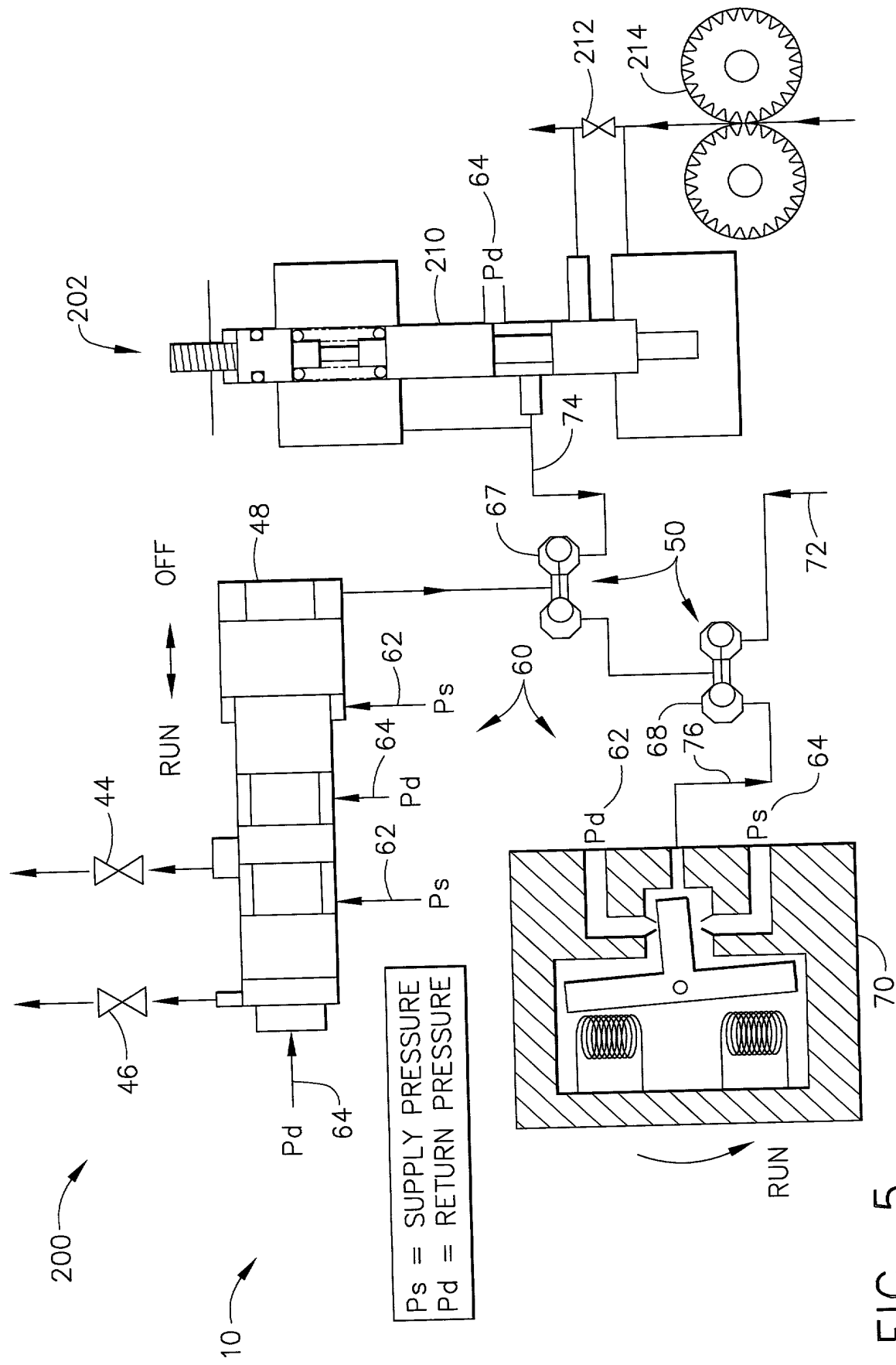


FIG. 4



**DECLARATION AND POWER OF ATTORNEY
FOR PATENT APPLICATION**

Docket Number
13DV13462

As a below named inventor, I hereby declare that:

My residence, post office address, and citizenship are as stated below next to my name.

I believe I am the original, first and sole inventor (if only one name is listed below) or an original, first and joint inventor (if plural names are listed below) of the subject matter which is claimed and for which a patent is sought on the invention entitled:

-----METHODS AND APPARATUS FOR ROTOR OVER-SPEED PROTECTION-----

the specification of which

☒ is attached hereto

OR

☐ was filed on _____ as United States Application Number or PCT International Application Number _____
and was amended on _____ (if applicable).

I hereby state that I have reviewed and understand the contents of the above identified specification, including the claims, as amended by any amendment specifically referred to above.

I acknowledge the duty to disclose information which is material to patentability as defined in Title 37 Code of Federal Regulations, §1.56.

I hereby claim foreign priority benefits under Title 35, United States Code §119 (a)-(d) or §365 (b) of any foreign application(s) for patent or inventor's certificate, or §365 (a) of any PCT international application which designated at least one country other than the United States of America, listed below and have also identified below any foreign application for patent or inventor's certificate, or of any PCT international application having a filing date before that of the application on which priority is claimed.

PRIOR FOREIGN APPLICATION(s)

Priority Claimed

☐ Yes ☐ No

☐ Yes ☐ No

(Number) (Country) (Day/Month/Year Filed)

(Number) (Country) (Day/Month/Year Filed)

☒ Additional foreign application numbers are listed on a supplemental priority data sheet attached hereto.

I hereby claim the benefit under Title 35, United States Code §119 (e) of any United States provisional application(s) listed below.

☐ Additional provisional application numbers are listed on a supplemental priority data sheet attached hereto.

(Application Number) (Filing Date)

I hereby claim the benefit under Title 35, United States Code §120 of any United States Application(s), or §365 (c) of any PCT international application designating the United States of America, listed below and, insofar as the subject matter of each of the claims of this application is not disclosed in the prior United States or PCT International application in the manner provided by the first paragraph of Title 35, United States Code §112, I acknowledge the duty to disclose information which is material to patentability as defined in Title 37, Code of Federal Regulations §1.56 which became available between the filing date of the prior application and the national or PCT international filing date of this application.

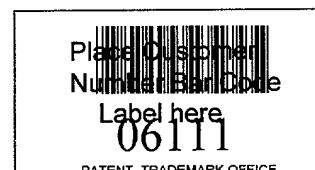
(Application Number) (Filing Date) (Status - patented, pending, abandoned)

(Application Number) (Filing Date) (Status - patented, pending, abandoned)

I hereby appoint the registered practitioners associated with Customer Number 006111 to prosecute this application and to transact all business in the Patent and Trademark Office connected therewith.

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CUSTOMER NUMBER: 006111

I hereby declare that all statements made herein of my own knowledge are true and that all statements made on information and belief are believed to be true; and further that these statements were made with the knowledge that willful false statements and the like so made are punishable by fine or imprisonment, or both, under Section 1001 of Title 18 of the United States Code and that such willful false statements may jeopardize the validity of the application or any patent issued thereon.

SOLE OR FIRST INVENTOR:

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First Name

Middle Name

Last Name

Signature: Kail Lester Linebrink

Date

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Full name: _____

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Middle Name

Last Name

Signature: _____

Date

Residence: _____

City and State

Citizenship: _____

Post Office Address: _____